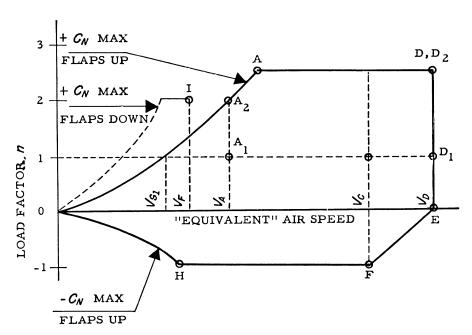
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- (iv) In cases where the flight deck pitch control motion may be affected by inputs from systems (for example, by a stick pusher that can operate at high load factor as well as at 1g), then the effects of those systems shall be taken into account.
- (v) Airplane loads that occur beyond the following times need not be considered:
- (A) For the nose-up pitching maneuver, the time at which the normal acceleration at the center of gravity goes below 0g;
- (B) For the nose-down pitching maneuver, the time at which the normal acceleration at the center of gravity goes above the positive limit load factor prescribed in §25.337;
 - (C) tmax.

§25.333 Flight maneuvering envelope.

- (a) General. The strength requirements must be met at each combination of airspeed and load factor on and within the boundaries of the representative maneuvering envelope (V-n diagram) of paragraph (b) of this section. This envelope must also be used in determining the airplane structural operating limitations as specified in §25.1501.
 - (b) Maneuvering envelope.



[Doc. No. 5066, 29 FR 18291, Dec. 24, 1964, as amended by Amdt. 25-86, 61 FR 5220, Feb. 9, 1996]

§25.335 Design airspeeds.

The selected design airspeeds are equivalent airspeeds (EAS). Estimated values of V_{S0} and V_{S1} must be conservative.

- (a) Design cruising speed, V_C . For V_C , the following apply:
- (1) The minimum value of V_C must be sufficiently greater than V_B to provide for inadvertent speed increases likely to occur as a result of severe atmospheric turbulence.
- (2) Except as provided in §25.335(d)(2), V_C may not be less than $V_B+1.32~U_{REF}$ (with U_{REF} as specified in §25.341(a)(5)(i)). However V_C need not exceed the maximum speed in level flight at maximum continuous power for the corresponding altitude.
- (3) At altitudes where V_D is limited by Mach number, V_C may be limited to a selected Mach number.
- (b) Design dive speed, V_D . V_D must be selected so that V_C/M_C is not greater